

Airworthiness Directive Schedule

Aeroplanes

Eagle X-TS and 150 Series

30 August 2007

The date above indicates the amendment date of this schedule.

New or amended ADs are shown with an asterisk *

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DCA/EAGLE/1 Rudder Cable Pulleys – Inspection and Modification

Applicability: X-TS 150 and 150B, S/N 001 through 030, not modified per Eagle SB 1076.

Requirement: To prevent fraying of the rudder cable, accomplish the following:-

1. Inspect rudder cables in the vicinity of the under-seat pulleys, per Eagle SB 1059.
2. Modify the under-seat rudder cable pulley installation and reset rudder cable tension, per Eagle SB 1076.
(Australian AD/X-TS/2 refers)

Compliance:

1. At intervals not to exceed 50 hours TIS.
2. Within next 100 hours TIS, or by 31 December 2000, whichever occurs first.

Effective Date: 24 February 2000

DCA/EAGLE/2 Canard Rear Spar - Reinforcement

Applicability: Models X-TS 150 and 150B S/N 001 through 003 and 005 through 030.

Requirement: To provide additional canard rear spar bracket strength to enable the aircraft to meet certification requirements, modify the canard rear spar bracket per Eagle SB 1074.
(Australian AD/X-TS/3 refers)

Compliance: Within next 100 hours TIS.

Effective Date: 24 February 2000

DCA/EAGLE/3 Throttle Torque Tube and Throttle Arm - Modification

Applicability: Models X-TS 150 and 150B S/N 001 through 021.

Requirement: To prevent failure of the co-pilot's throttle control, modify the throttle torque tube and throttle arm per Eagle SB 1067 Revision 1.
(Australian AD/X-TS/4 refers)

Compliance: By 27 July 2001

Effective Date: 27 July 2000

DCA/EAGLE/4A Flap Hinge Support Bracket - Inspection

Applicability: Model XTS-150 and 150 B, not modified in accordance with Eagle Aircraft SB1109.

Requirement: To prevent failure of the inner flap attachment which may result in asymmetric flap deployment and/or inability to lower or raise the flaps, accomplish the following;

1. Visually inspect the gusset weld area for cracks, see figure 1 for typical crack location. If there is any cracked, lifted or missing paint in the area of the weld, or a crack is suspected, then perform detailed inspection per Part 2 of this AD before further flight. If bracket integrity is assured, no further action is required.
2.
 - a. Remove paint in area of weld.
 - b. Visually inspect the weld area for cracks using a bright light and 10X magnifying glass.
 - c. If bracket cannot be verified free of cracks, then inspect using Fluorescent Penetrant Inspection. Any cracked brackets must be replaced or repaired in accordance with approved data before further flight.

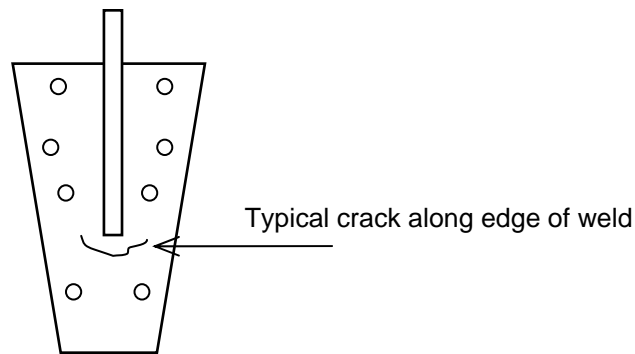


Figure 1

3. Embodiment of Eagle Aircraft (M) SB 1120 constitutes terminating action for the requirements of this AD.

Note 1: Eagle Aircraft (M) Sdn.Bhd. Service Bulletins may be obtained from CTRM Aviation Sdn. Bhd. PO Box 1028 Pejabat Pos Besar Melaka, Malaysia. Ph (606) 317 4105, Fax (606) 317 7213, email: product.support@ctrmeagel.com
(Malaysian CAM AD 001-01-2004 R1 refers)

Compliance: At each daily inspection of the aircraft. Inspection requirement to be noted on aircraft tech log.

Note 2: The daily repetitive inspections may be accomplished by the pilot in accordance with CAR Part 43, Appendix A. The pilot must be trained and authorised (Part 43, Subpart B refers) and certification must be provided (Part 43, Subpart C refers).

Effective Date: 26 January 2006

DCA/EAGLE/5 Main Undercarriage Attachment Bracket - Inspection

Applicability: Model X-TS, X-TS 150, and 150B.

Requirement: To prevent failure of the undercarriage attachment, inspect per Eagle Aircraft SB 1123. If the adjoining plates on the undercarriage attachment bracket are not welded, replace the bracket before further flight.
(Malaysia DCA CAM AD 001-08-2004 refers)

Compliance: Before further flight.

Effective Date: 21 August 2004

DCA/EAGLE/6 Co-pilot Rudder Pedals

Applicability: Model X-TS, X-TS150 and 150B

Requirement: To prevent binding of the rudder pedals caused by excessive wear of the pivot, accomplish the following:

1. Remove the co-pilot rudder pedal assembly P/N 2720D07-02 and accomplish Eagle Aircraft MSB 1095. Install placard as detailed in the SB or,
2. Install new bush P/N 2720D08-39 and rework co-pilot rudder pedal assembly P/N 2720D07-02 in accordance with Eagle Aircraft SB 1096 or,
3. Replace co-pilot rudder pedal assembly with new assembly P/N 2720D07-10 in accordance with Eagle Aircraft SB 1097.
(Malaysian AD 002-10-2004 refers)

Compliance: Accomplish either 1, 2, or 3 before further flight unless already accomplished.

Effective Date: 25 November 2004

*** DCA/EAGLE/7 Flap Hinges & Support Brackets – Inspection and Replacement**

Applicability: Model X-TS, XTS-150 & 150B aircraft, S/N 01 through to 42 and M1001 through to M1003.

Model 150B aircraft, S/N 43, 44 and M1005 onwards.

Requirement: To prevent failure of the flap hinges and support brackets due to the possibility of corrosion resulting in loss of aircraft control, accomplish the instructions and corrective actions specified in CTRM Aviation Sdn. Bhd. Service Bulletin No. 1126.
(Malaysian AD CAM AD 001-07-2007 refers)

Compliance: Within the next 50 hours TIS or by 30 September 2007, whichever occurs sooner.

Effective Date: 30 August 2007